of the gas turbine engine and said primary flow includes at least a combustion core gas exhaust flow sourced from a turbine section of the gas turbine engine and a flow from a compressor section.

8. The nozzle assembly of claim **1**, wherein the flexible panel is formed at least partially of an organic matrix composite.

9. The nozzle assembly of claim **1**, wherein the flexible panel further comprises a stiffening stay operative to maintain a profile of a throat area.

10. The nozzle assembly of claim **1**, wherein the flexible panel further comprises a support structure operative to support the flexible panel.

11. The nozzle assembly as recited in claim **1**, wherein said secondary flow is selectively sourced only from a fan section of the gas turbine engine and said primary flow includes at least a combustion core gas exhaust flow sourced from a turbine section of the gas turbine engine and a flow from said fan section.

12. The nozzle assembly as recited in claim **1**, wherein said secondary flow is selectively sourced only from a fan section of the gas turbine engine and said primary flow includes at least a combustion core gas exhaust flow sourced from a turbine section of the gas turbine engine and a flow from a compressor section.

13. A nozzle assembly for a gas turbine engine, the nozzle assembly comprising:

a two-dimensional secondary nozzle having a throat area and an exit area;

a flexible panel operative to variably alter at least one of the throat area and the exit area to regulate a secondary flow from the two-dimensional secondary nozzle, said secondary flow different than a combustion core gas exhaust flow sourced from a turbine section of the gas turbine engine;

a cavity on a non-gas side of said flexible panel; and

an actuator located in said cavity, said actuator configured to deflect said flexible panel, said actuator oriented in a direction that is non-parallel to an axis of rotation of the gas turbine engine.

14. The nozzle assembly of claim **13**, further comprising:

a stiffening rib operative to deflect the flexible panel such that at least one of the throat area and the exit area is altered.

15. The nozzle assembly of claim **14**, wherein said actuator is connected to the stiffening rib and is operative to deflect the flexible panel by positioning the stiffening rib.

16. The nozzle assembly of claim **13**, wherein the panel is operative to symmetrically affect the secondary flow with respect to yaw.

17. The nozzle assembly of claim **13**, wherein the nozzle assembly further comprises a support structure operative to support the flexible panel.

18. The nozzle assembly of claim **13**, wherein said cavity is operative to reduce an actuation load associated with positioning of the panel by reducing pressure imbalances on the non-gas side and a gas path side of the flexible panel.

19. A gas turbine engine comprising:

a secondary flow duct with a two-dimensional secondary nozzle to communicate a secondary flow therethrough;

a primary flow duct with a two-dimensional primary nozzle to communicate primary flow therethrough, said two-dimensional primary nozzle adjacent to said two-dimensional secondary nozzle, wherein said primary flow includes at least a combustion core gas exhaust flow sourced from a turbine section of the gas turbine engine, and wherein said secondary flow is different than said combustion core gas exhaust flow;

a flexible panel adjacent said two-dimensional secondary nozzle, said flexible panel operable to selectively define a range of positions to regulate said two-dimensional secondary nozzle;

a cavity on a non-gas side of said flexible panel; and

an actuator located in said cavity, said actuator configured to deflect said flexible panel, said actuator oriented in a direction that is non-parallel to an axis of rotation of the gas turbine engine.

20. The gas turbine engine of claim **19**, further comprising a stiffening rib configured to deflect the flexible panel.

21. The gas turbine engine of claim **20**, wherein said actuator is connected to the rib and is configured to operatively rotate the stiffening rib and deflect the flexible panel in a throat area and an exit area of the two-dimensional secondary nozzle.

22. The gas turbine engine of claim **20**, wherein the two-dimensional secondary nozzle is a third stream exhaust nozzle assembly.

23. The gas turbine engine of claim **20**, wherein further comprising a stiffening stay configured to maintain a profile of a throat area of said two-dimensional secondary nozzle.

24. The gas turbine engine of claim **19**, wherein said flexible panel is located on one side of said two-dimensional secondary nozzle opposite said two-dimensional primary nozzle.

25. The gas turbine engine of claim **19**, wherein said two-dimensional secondary nozzle is downstream of said two dimensional primary nozzle.

26. The gas turbine engine of claim **19**, wherein said two-dimensional secondary nozzle is adjacent to said two-dimensional primary nozzle.

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